



A Mahindra Aerospace Company

PO Box 881, Morwell, Victoria 3840, Australia
Ph + 61 (0) 3 5172 1200
Fax + 61 (0) 3 5172 1201
www.gippsaero.com.au

SB-GA8-2021-207

Issue 1

OPTIONAL

Service Bulletin

Subject:

Installation of external LED lights

Applicability:

This Service Bulletin is applicable to the aircraft identified in Table 1.

Table 1 – Applicability

AIRCRAFT	SERIAL NUMBER(s)
GA8	All
GA8-TC 320	All

Amendments:

Issue 1: Initial Issue

Background:

This Service Bulletin provides instructions for the replacement of external filament lights located in right wing, left wing, and vertical fin with 14V Whelen LED lights.

The Whelen light models PLED1L and PLED1T are FAA approved for aviation use as a landing and taxi lights. Refer STC SA00344BO. These LED lights are also approved as PMA parts as "drop in" replacements on GA8 models.

This Service Bulletin installation is equivalent to external LED lighting installed in accordance with GippsAero Engineering Release ER-GA8-9633227.

Compliance:

The accomplishment instructions contained within this Service Bulletin are optional and may be incorporated at the Operator's, Owner's or Maintenance Provider's discretion.

Weight and Balance:

Refer Part C

Electrical Load Analysis:

The effect of this Service Bulletin's incorporation on the aircraft's electrical load analysis is shown in Table 2 & Table 3. Subtract the electrical load for those components removed and replaced with LED lighting and amend the aircraft's ELA.

Table 2 – Electrical Load Analysis

COMPONENT	NOMINAL CURRENT @ 14V	CIRCUIT BREAKER LABEL	BUS	PHASES OF FLIGHT
2 x Whelen PLED1L Landing Lights	2.8 A (combined)	LANDING	1	Take-off Landing
2 x Whelen PLED1T Taxi Lights	2.8 A (combined)	TAXI	2	Take-off Landing Taxi
2 x Whelen LED OR6001 Wing Nav Lights	1.0 A (combined)	NAV (Position)	2	All
	2.0A (combined)	STROBE (ACL)	2	
1 x Whelen LED OR5001V Tail Nav Light	0.2A	NAV (Position)	2	All
	0.8A	STROBE (ACL)	2	

The aircraft's electrical load analysis shall be updated to include this information.

Table 3 Existing GA8 Filament/Strobe

COMPONENT	NOMINAL CURRENT @ 14V	BUS
2 x GA8-334013-15 Landing Lights	9.4 A	1
2 x GA8-334013-15 Taxi Lights	9.4 A	2
2 x Whelen 01-0770054-00 Position/Anti-collision Light with Power Supply unit 01-0770006-09.	7.2 A	2
Whelen 01-0770024-00 Position/Anti-collision Light with Power Supply unit 01-0770006-09.	2.4 A	2
2 x Whelen W1285P 14 Position Light Wing (Red), 1 X Whelen A555A 14 Position Light Tail (White)	4A 2A	Bus2/ Main
1 x 150-0011 Aeroflash Flasher Unit and 2 x 151-0002 Aeroflash Beacon Light Assemblies.	(1)	Bus2/ Main

(1) Approx. 9A. Measure actual electrical load before removal.

Approval:

The airframe and/or electrical system modification/repair described in this Service Bulletin has been approved pursuant to Australian Civil Aviation Safety Regulation 21.095 (1998). GippsAero Reference GAE11#2761.

Parts:

The following parts are required to accomplish this Service Bulletin and are available as kit Part Number SB-GA8-2021-207-01.

Table 4 - Parts

ITEM	PART No.	DESCRIPTION	QTY
1	01-0771733-01	OR6001G WINGTIP PTA,12V (GREEN)	1
2	01-0771733-02	OR6001R WINGTIP PTA,12V (RED)	1
3	01-0771774V01	OR5001V ORION500 TAIL PTA LIGHT WHITE, 12V	1
4	01-0771833-10	WHELEN P36P1L LANDING LIGHT	2
5	01-0771833-15	WHELEN P36P1T TAXI LIGHT	2
6	1-480426-0	CONN CAP HSG 4 POS COMMERCIAL MATE-N-LOK	2
7	1-480426-0	CONN PIN HOUSING 4 POS (P58)	1
8	1-480704-0	CONN PLUG HOUSING 6 POS MATE-N-LOK (P39A)	1
9	1-480705-0	CONN CAP HOUSING 6 POS MATE-N-LOK (J39, J39A)	2
10	1-480709-0	CONN CAP HOUSING 12 POS MATE-N-LOK (J01, J02)	2
11	163305-2	CONTACT PIN 20-18AWG CRIMP	8
12	163305-2	CONTACT PIN 20-17AWG CRIMP	4
13	350689-1	CONTACT CRIMP SKT 24-18 AWG MATE-N-LOK	33
14	350690-1	CONTACT PIN 24-18 AWG CRIMP MATE-N-LOK	5
15	E1000-3	NYLON TUBE 3/16 O.D.	20"
16	GA8-246045-017	WIRE KIT FOR OVERHEAD PANEL ELECTRICAL LED LIGHTS INSTALLATION	1
17	GA8-246045-019	WIRE KIT FOR FUSELAGE LED INSTALLATION	1
18	GA8-246072-011	ELECTRICAL HARNESS VERTICAL FIN	1
19	GA8-246073-011	ELECTRICAL HARNESS LEFT WING	1
20	GA8-246073-015	SINGLE WIRE FUEL RETURN	1
21	GA8-246073-017	SINGLE WIRE PITOT HEAT RETURN	1
22	GA8-246074-011	ELECTRICAL HARNESS RIGHT WING	1
23	GA8-246074-013	SINGLE WIRE STALL RETURN	1
24	GA8-246074-015	SINGLE WIRE FUEL RETURN	1
25	GA8-334023-045	LED CLAMPING PLATE	4
26	GA8-334028-021	NAVIGATION LAMP BASE (OR6001)	2
27	LHMS-55-D	LIGHTENING HOLE MOUNT	2
28	M81824/1-2	SPLICE INSULATED AWG 20-16 BLUE 105C	20
29	M83519/2-7	SOLDER SLEEVE 0.105/0.055 LEADED 22AWG	4
30	M83519/2-8	SOLDER SLEEVE 0.170/0.085 LEADED 22AWG	4
31	M83519/2-9	SOLDER SLEEVE 0.235/0.130 LEADED 22AWG	8
32	MS25036-149	TERML LUG INSUL RING 22-18AWG #8 DIA RD	10
33	NAS548-P8-16	SCREW - FLAT HEAD	6
34	PAF28	CONDUIT, PLASTIC, FLEXIBLE, 28MM	3.5m
35	TLPD321BS	RIVET, POP	30
36	TLPD419BS	RIVET, POP	12
37	TLPD424BS	RIVET, POP	2
38	TLPD435BS	RIVET, POP	5
39	W31X2M1G-5	CIRCUIT BREAKER SWITCH 5A	4

The following consumables are required to accomplish this Service Bulletin.

Table 5 - Consumables

ITEM	PART No.	DESCRIPTION	QTY
C1	RTV 747	SILICONE SEALANT (OR EQUIVALENT NEUTRAL CURE SEALANT)	1

Parts Availability:

New parts can be obtained directly from GippsAero.

Tel: +61 (0)3 5172 1200

Fax: +61 (0)3 5172 1201

Email: PARTS@gippsaero.com.au

Labour:

32 man hours should be allocated for completing the work detailed in this Service Bulletin. This time does not include set up etc.

Warranty:

Optional modification warranty is not applicable.

Accomplishment Instructions:

The following instructions are applicable to the left hand side (LHS) of the aircraft; the right hand side (RHS) is opposite, unless noted otherwise.

WARNING:

IT IS THE RESPONSIBILITY OF ALL PERSONNEL TO ENSURE WORK HEALTH AND SAFETY REQUIREMENTS ARE MET AT ALL TIMES. ALL PERSONNEL MUST COMPLY WITH ALL WORK HEALTH AND SAFETY REQUIREMENTS AS DEFINED OR RECOMMENDED BY:

- EQUIPMENT OEM INSTALLATION AND OPERATION MANUALS;
- AIRCRAFT MAINTENANCE AND OPERATION MANUALS;
- ASSOCIATED AIRCRAFT MODIFICATION INSTRUCTIONS;
- RELEVANT NAA REGULATIONS AND ADVISORY DOCUMENTATION;
- ORGANISATION MANUALS, INCLUDING NAA ENDORSED OPERATIONAL AND MAINTENANCE MANUALS; AND
- RELEVANT LOCAL, STATE AND FEDERAL GOVERNMENT REQUIREMENTS.

WARNING:

READ THE APPLICABLE MATERIAL SAFETY DATA SHEET (MSDS) FOR ANY MATERIAL/CONSUMABLE USED DURING THE ACCOMPLISHMENT OF THIS SERVICE BULLETIN AND EMPLOY ANY RECOMMENDED PERSONAL PROTECTIVE EQUIPMENT (PPE) CONTAINED THEREIN.

NOTE:

Unless otherwise specified, reference to the GA8/GA8-TC 320 Service Manual and FAA Advisory Circular (AC) 43.13-1B should be made when carrying out the procedures prescribed in this Service Bulletin. In case of a discrepancy between the Service Manual and the AC, the Service Manual takes precedence.

Part A – Installation

Preparation & General notes

1. The installation of wiring is to be performed in accordance with FAA Advisory Circular 43.13-1B, Chapter 11 - Acceptable Methods, Techniques, and Practices - Aircraft Inspection and Repair - for wiring purpose. Carry out grounding and bonding checks on all disturbed electric wiring.
2. Refer to GA8 Service Manual, Chapter 24 (Electrical Power) for safety precautions & procedures to be adhered to when performing any maintenance or modification to the electrical system.
3. Obsolete wiring may be removed or capped and stowed as appropriate.
4. Where equipment is removed, restore or repair the structure in accordance with FAA AC43.13-1B
5. Prior to installation carry out a baseline EMC test (refer Part B) and record if there is any electrical interference.
6. Ensure the aircraft is safe for maintenance.
7. Turn both Master switches to the OFF position.
8. Pull all under floor circuit breakers.
9. Disconnect the Battery and remove ground power.
10. In the electrical overhead panel, turn OFF the Landing Light, Taxi Light, Cabin Light, Wing Light, Nav Light and Strobe Light circuit breakers.
11. Typical Mate-N-Lok pin numbering is shown in Figure 1. Refer TE Connectivity Mate-N-Lok AMP Connectors datasheets for additional information on use.

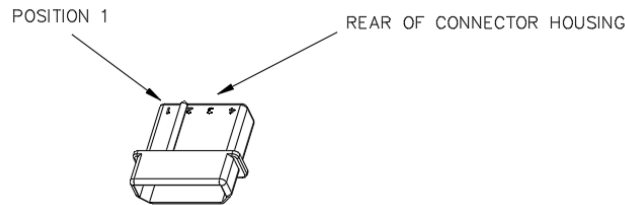


Figure 1 : Mate-N-Lok pin numbering

Wing Installation

12. Remove the existing LH & RH landing light assemblies.
13. Remove the LH & RH wing tips.
14. Remove and discard the existing strobe light power supply units.
15. Gain access to the existing wiring by removing
 - wing leading edge access panels
 - forward wing root fairings
 - fuel float sensor access panels.
16. Replace the existing looms with the following wires and harness assemblies. Retain and incorporate any additional wiring required additional modifications.
 - 16.1. LH & RH Wing Electrical harness's (Item 19 & Item 22)
 - 16.2. LH & RH Wing Single wire fuel return (Item 20 & Item 24)
 - 16.3. LH Wing Single wire pitot heat return (Item 21)
 - 16.4. RH Wing single wire stall return (Item 23).
17. Conduct a continuity check of the added aircraft wiring. Refer Figure 10. Carry out a bonding and grounding check on all new and disturbed wiring in accordance with FAA AC43.13-1B

Landing/Taxi Lights

18. Choose the appropriate replacement light assembly. Install the landing light PLED1L (Item 4) into the inboard position and the taxi light PLED1T (Item 5) into the outboard position of the landing light assemblies.
19. Inspect the landing light assemblies for any signs of abnormal wear or damage.
20. Remove and retain the 4 screws holding the clamping plate and existing lamp in place. Ensure the clamping plate (GA8-334023-35) doesn't fall from the lamp assembly before it is ready to be removed.
21. Remove the lamp and clamping plate from the receptacle and discard both.
22. Install the rubber gasket included with the LED Light around the LED Light assembly. A light application of a common hand soap on the rubber may aid in its fit into the light housing.
23. Place the LED light into the receptacle. Proper orientation of the lamp is necessary for beam orientation to be left and right of centreline while in operation. For a Taxi light, the 'lines' of the inner optic should be orientated vertically to produce a horizontal light beam.
24. Install the LED clamping plate (Item 25). Care should be taken to tighten the screw only to the point the retainer does not turn with hand pressure. Over-tightening may stress the polycarbonate lens of the LED Light.
25. Connect wires to the light assemblies Refer Figure 2, noting the terminal markings for positive and negative. Refer Figure 10 attach wiring to the terminal posts of the LED Lights.
26. Refit the LH & RH landing light assemblies.

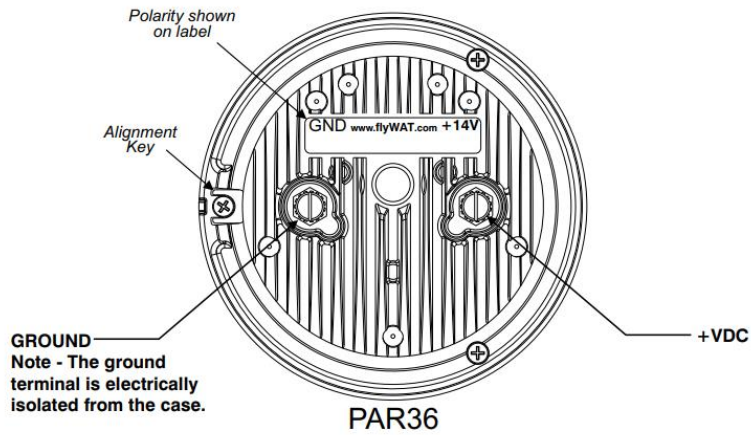
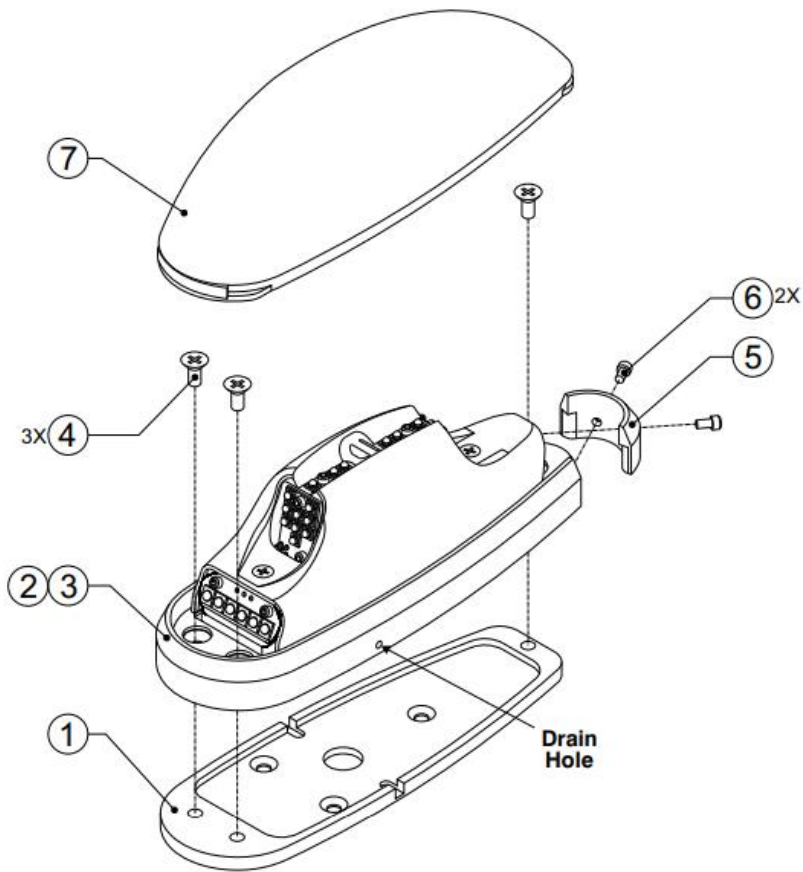


Figure 2 : LED Light Connections

Nav Lights

27. Remove the existing navigation lights and mounting pads from the wing tips.
28. Install the LED Position/Anti-collision lights (Items 1 & 2) to the wing tip navigation light pads as below
 - 28.1. Choose the appropriate light assembly. Use red light for port side, and green for starboard.
 - 28.2. Cut supplied nav light wires to 10.5" in length. Cut 8.75" tubing (Item 15) and install over wires. Crimp pin contacts (Item 11). Refer Figure 10 and insert wires into connector housings P55 & P56 (Item 6).
 - 28.3. Refer Figure 3, carefully remove the #2 cap head screws and lens retainer. Remove the lens from the light assembly by lifting the rear of the lens approximately 1/2". Now slide the lens rearward approximately 1/2" and lift upwards to remove. CAUTION! Do not touch the LED lens surface with either fingers or sharp objects. This could soil and/or damage the lens and affect the optical performance of the LEDs. Remove the 3 black phillips head screws securing the baseplate to the light assembly. Remove baseplate.
 - 28.4. Apply a thin layer of sealant (Item C1) between the wing tip and the mounting pads. Secure the new navigation light baseplate and mounting pads (Item 26) to the wingtip using screws (Item 33).
 - 28.5. Re-install the light assembly on to the baseplate and ensure that all leads are clear of any obstructions and secured as required. Note that proper orientation is achieved with the drain hole down.
 - 28.6. Install lens in the reverse order as removal, return the lens retainer to its installed location, re-insert #2 cap head screw and tighten firmly. Confirm proper gasket fit.
 - 28.7. Waterproof the light base to the aircraft. Apply sealant (C1) around any open area where water could get in. Do not cover the drain hole.
29. Connect the wingtip wiring and refit the wingtip assemblies.



QTY	QTY	ITEM	PART NUMBER	DESCRIPTION
	*		01-0771733-01	OR6001G Wingtip PTA, 12V (Green)
*			01-0771733-02	OR6001R Wingtip PTA, 12V (Red)
1	1	1	06-171708-001	Baseplate
	1	2	-	OR6001 Assembly (GREEN)
1		3	-	OR6001 Assembly (RED)
3	3	4	14-0050581B03	Screws, 4-40 x 5/16 P100FH MS24693-3B
1	1	5	19-171730-00	Retainer, Lens
2	2	6	14-026A36-04M	Screws, 2-56 x 1/4 Socket HP Cap
1	1	7	02-0371773-30	Lens, Clear, Hardcoated

Figure 3 : Wingtip light assy.

Vertical Fin Installation

- 30. If required to gain sufficient access, remove the fin. Refer to the aircraft service manual §55-30-00.
- 31. For aircraft fitted with strobe lights, remove the Vertical Fin Anti-Collision Lights the Power Supply located inside the tailcone at station 206.
- 32. For aircraft fitted with beacons, remove the beacons and install a patch repair in accordance with FAA AC43.13-1B. Remove the tail nav light. Remove the beacon flasher units and discard.
- 33. Remove the Fin Tip Beacon Box P/N GA8-553021-101 by drilling out the eight rivets (per side) refer in Figure 4.

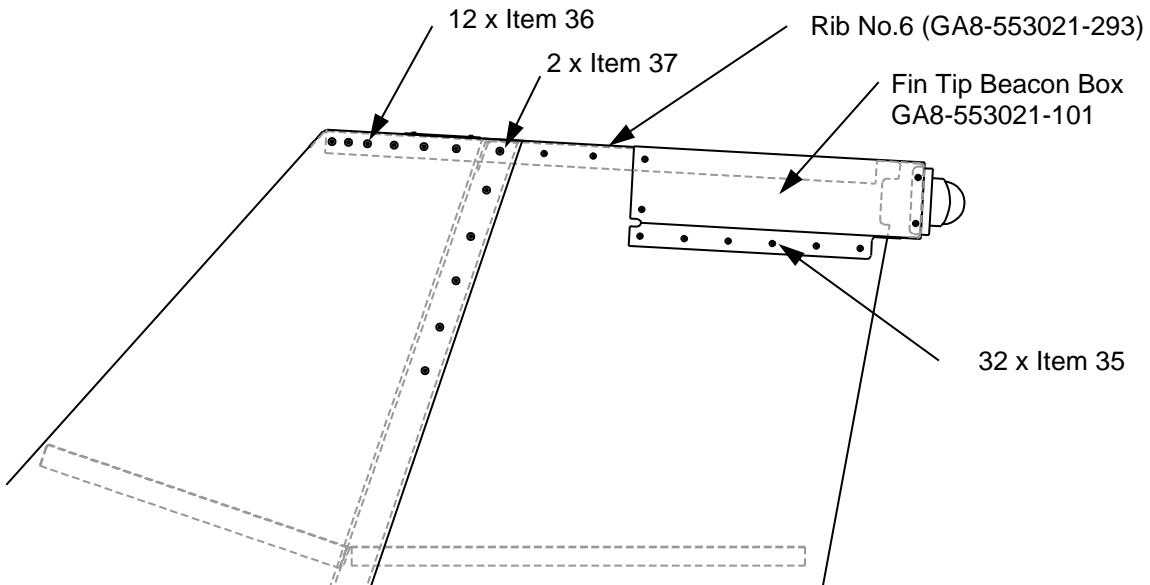


Figure 4 : Fin Tip Alteration

- 34. Remove and retain Rib No.6 (GA8-553021-293) at top of the Vertical Fin (refer Figure 6).
- 35. Modify the Fin by enlarging the original strobe light cut-out. Remove the material in the hatched region shown in Figure 6. Note that all dimensions are in inches, tolerances are ± 0.032 " and sharp corners should be suitably radiused. Trim Rib No.6 to suit cutout.

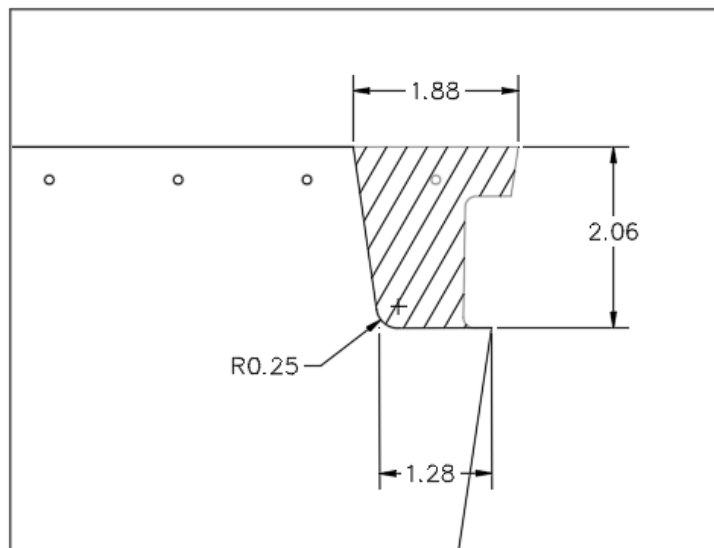


Figure 5 : Fin cut out

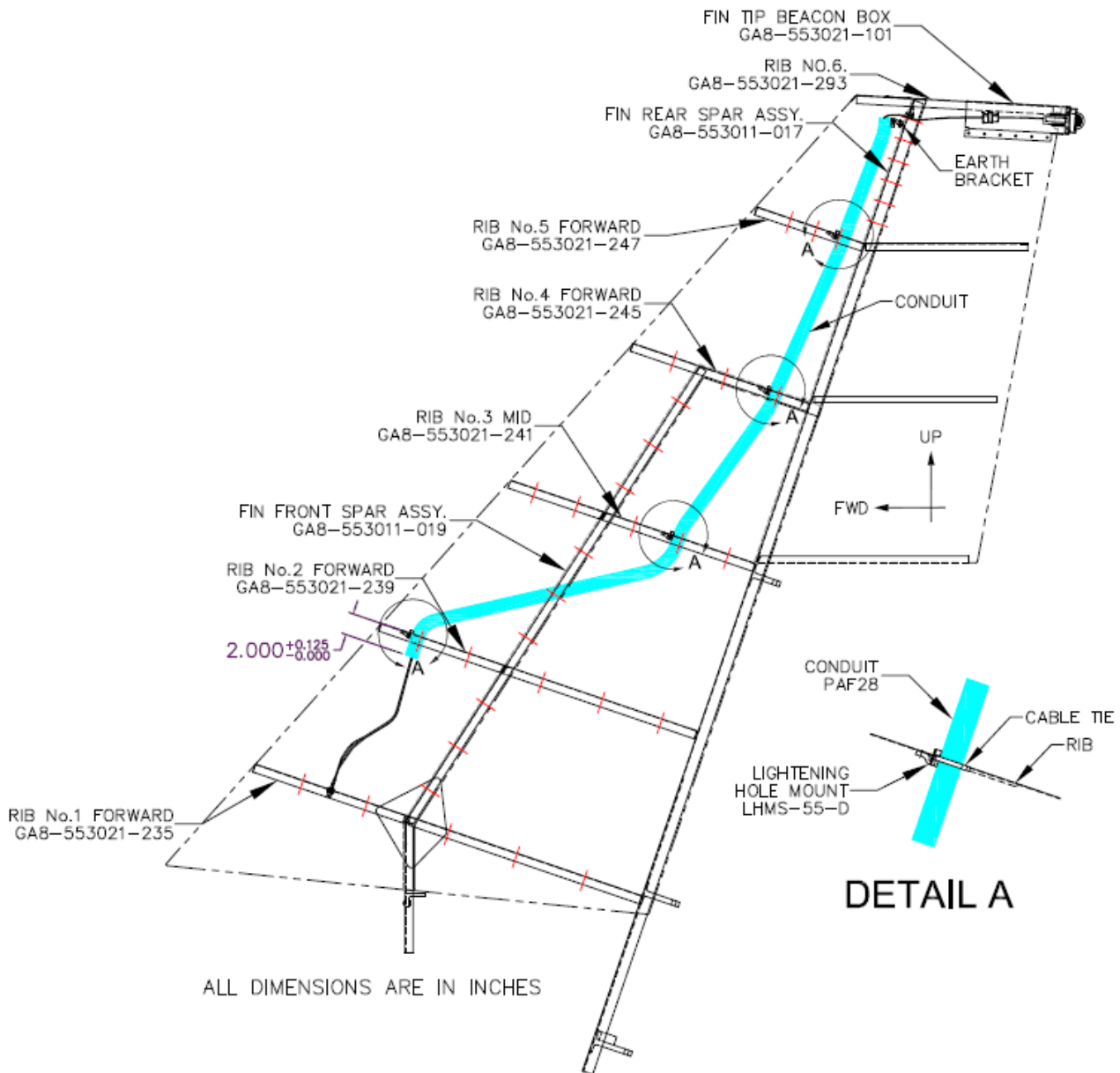


Figure 6 : Fin wire routing

36. Install the Electrical Harness (Item 18) through the conduit (Item 34). Pass the conduit through the Fin as shown in Figure 6. Note that the red lines indicate the axis of lightning holes inherent in the respective components. Mounts at ribs 3 & 4 may be omitted if they are inaccessible.
37. For each of the locations identified by Detail A in Figure 6, the Lightning Hole Mounts (Item 27) are to be fastened to the identified rib components using a pop rivet (Item 38) (drill Ø4.1mm hole).
38. An additional Lightning Hole Mount is to be secured to the existing Earth Bracket (GA8-553021-107) as shown in Figure 7.

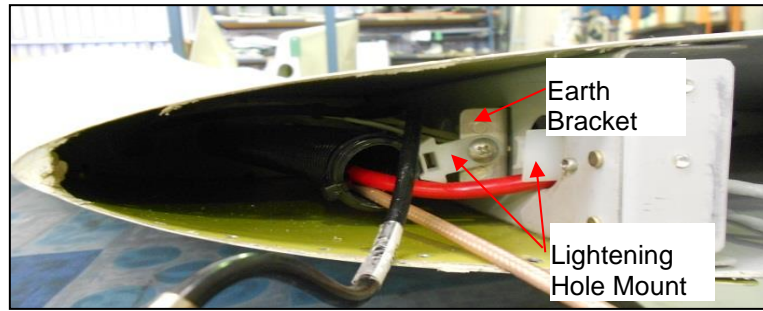


Figure 7 : Fin Tip

39. Fasten the conduit to the Lightning Hole Mounts as per Figure 6. Note that the lower end of the conduit should extend 2.000" (+0.125", -0.000") below the upper surface of Rib No.2 Forward (GA8-553021-239) component.
40. An additional Lightning Hole Mount (Item 27) is to be installed to the upper Fin Rear Spar Assembly (GA8-553011-017) lightning hole using a pop rivet (Item 38) (Ø4.1mm hole) as shown in Figure 7. Using a cable tie, secure wiring required to pass through this lightning hole to the Lightning Hole Mount. Ensure that the wiring does not chafe around the lightning hole.
41. Refit Rib No.6 (GA8-553011-293) and Fin Tip Beacon Box (GA8-553011-101) using pop rivets (12 x Item 36, 2 x Item 37 and 30 x Item 35), refer Figure 4.
42. Install 4-way plug J58 (supplied with Item 18). Conduct a continuity check of the added aircraft wiring.
43. Reconnect the Nav antenna coax or cap and stow as required.
44. Install the new LED Position/Anti-collision light (Item 3) to the fin tip beacon box as below.
 - 44.1. Cut supplied tail light wires to appropriate length. Crimp pin contacts (Item 12). Insert wires into connector housing P58 (Item 7) refer Figure 10.
 - 44.2. Connect the light wiring to the tail loom in accordance with Figure 10.
 - 44.3. Refer Figure 8 Install the light assembly on to the aircraft (discard the supplied nuts). Ensure that all leads are clear of any obstructions and secured as required. Note that proper orientation is achieved with the drain hole down.
 - 44.4. Waterproof the light base to aircraft. Apply sealant (C1) around any open area where water could get in. Do not cover the drain hole or the back of the unit.
45. Refit the fin. Refer to the aircraft service manual §55-30-00.

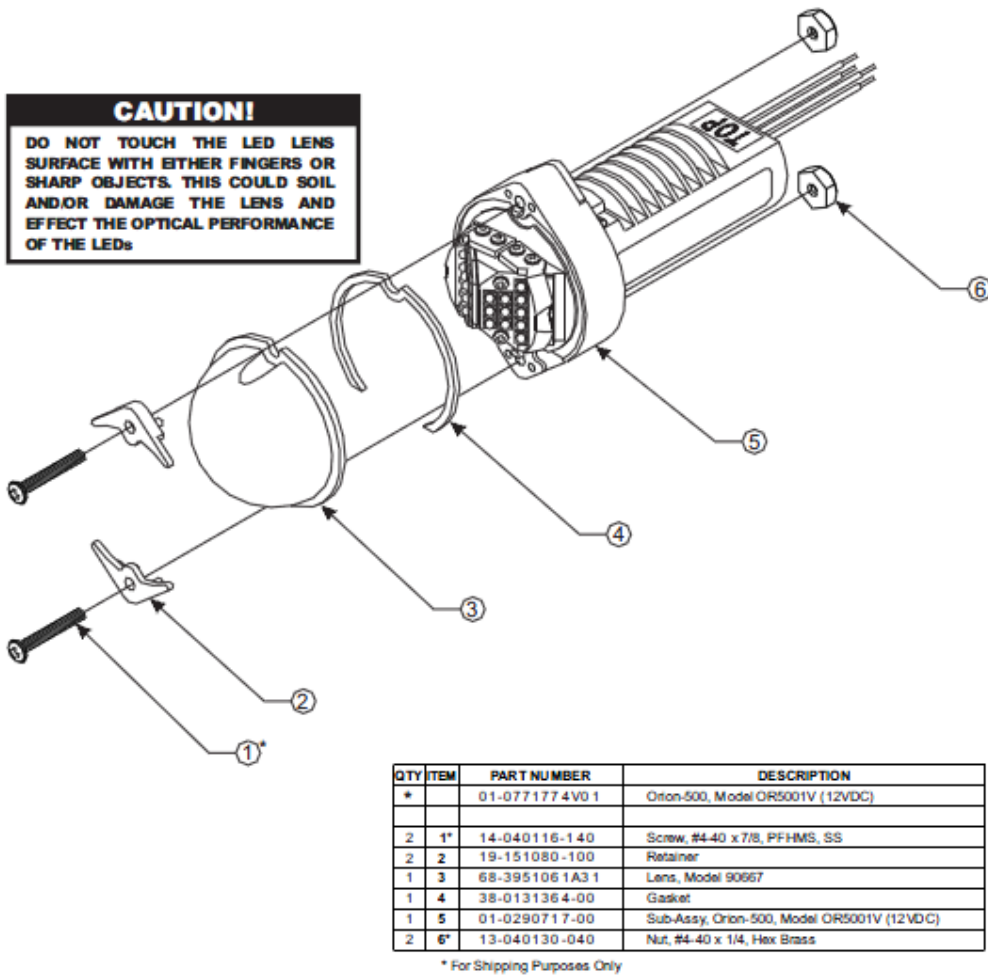


Figure 8 : Fin anti-collision and position light

Overhead Panel & Fuselage Wiring

- 46. Gain access to the overhead electrical panel.
- 47. Remove the existing 10A Landing Light, Taxi Light, Strobe light & Nav Light circuit breakers and discard. Install replacement 5A circuit breaker switches (Item 39)
- 48. If DC-DC converter is installed in the cabin roof at F.S.66.00 connect lighting grounds (Refer Figure 10 Note 1) to the earth point shown in Figure 9.
- 49. If no DC-DC

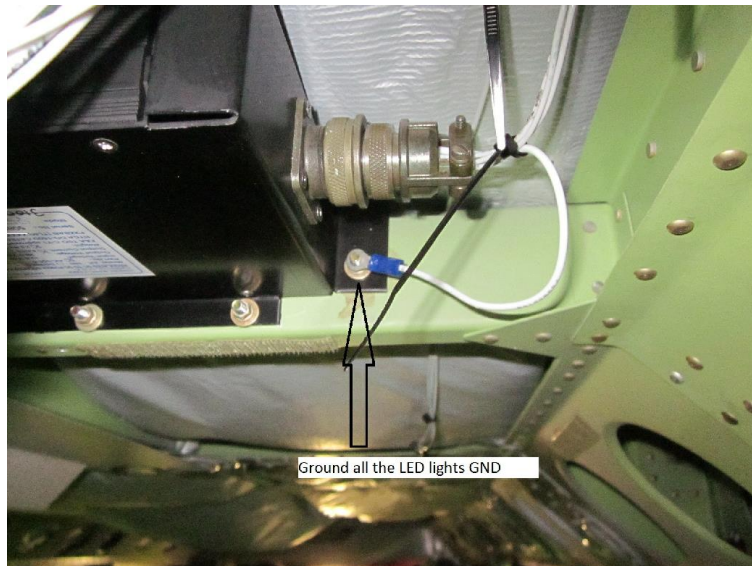


Figure 9 Overhead Panel Ground

- 50.
51. Modify the overhead panel & fuselage electrical wiring using wire kits (Item 16 & Item 17). Install connectors and pin/sockets listed in Table 6. Select appropriate solder sleeves and splices from Table 4 to complete the wiring in accordance with Figure 10.

Table 6 Connectors

Designation	Housing Item no.	Pins / Socket	
		Item no.	Qty
J01	10	13	12
J02	10	13	11
J39A	9	13	5
P39A	8	14	5
J39	9	13	5

52. Conduct a continuity check of the added aircraft wiring.
53. Re-secure the overhead electrical panel in place.

Part B – Testing

54. Reconnect the Aircraft Battery.
55. Close all the under-floor circuit breakers.
56. Apply external power.
57. Turn both Master switches to ON position.
58. Engage the Landing Light, Taxi Light, Cabin Light, Wing Light, Nav Light, Strobe Light circuit breakers one at a time and verify the functioning of their respective lights.
59. Align the new installed LED Landing Light & Taxi Light beams using the slotted holes provided.
60. Reinstall any removed access panels, fairings and interior linings.
61. Perform EMC check by cycling the source and observing the effect on the victim systems. LED lighting shall be evaluated as both the source and the victim. Complete the EMC Check Matrix in Table 7

The equipment shall not be the source of harmful conducted or radiated interference or adversely affect other equipment or systems installed in the airplane. With the equipment energized on the ground, individually operate other electrically operated equipment and systems on the airplane to determine that no significant conducted or radiated interference exists. Evaluate all reasonable combinations of control settings and operating modes. Operate communication and navigation equipment on at least one low, high and mid-band frequency. Make note of systems or modes of operation that should also be evaluated during flight. For airplane equipment and systems that can be checked only in flight, determine that no operationally significant conducted or radiated interference exists. Evaluate all reasonable combinations of control settings and operating modes.

Table 7 : EMC Check Matrix

System	Specific Item	Failure Indication	OK
Avionics Instruments	General	Blinking, Flickering, distortion	
	ASI	Change in displayed Airspeed	
	Altitude	Change in displayed Altitude	
	VSI	Change in Displayed Vertical Speed	
	Heading	Change in displayed Heading	
	Attitude	Change in displayed Attitude	

System	Specific Item	Failure Indication	OK
	COM	Noise, Squelch, Clicking	
	NAV	Noise, Squelch, Clicking, Unstable indication	
	GPS	Change in position, GPS Status - Position accuracy fields - Unstable position data. Check Satellite page for signal strength.	
	Audio System / Speaker	Audio Noise	
	XPDR / ADS-B	Inability to transmit	
	Traffic	Loss of data	
	Rad Alt	Change in displayed Altitude	
	Standby Compass	Change in displayed Heading	
Engine & Fuel Instruments	All	Erroneous Parameter display, Errors Displayed	
Instrument Back Lighting	All	Dimming, Flickering	
Lighting Electrical	Landing	Dimming, Flickering	
	Taxi	Dimming, Flickering	
	Nav	Dimming, Flickering	
	Strobe	Dimming, Flickering	
	Coaming	Dimming, Flickering	
	Courtesy	Dimming, Flickering	
	Map	Dimming, Flickering	
Electrical	Fuel Pumps	Unsteady flow	
	Warning lights	Pulsing, incorrect or erratic display	
	Pitot Heat	Temperature instability	
	Clock / CO Det	False warnings	
	Magnetos	Loss of ignition	
	ACU	Shutdown of Alternator, Loss of power or power control.	
	ACU (secondary) (if installed)	Shutdown of Alternator, Loss of power or power control.	

System	Specific Item	Failure Indication	OK
Power Distribution & Generation	Control Logic	Erratic operation of contactors or loss of power	
	Generator	Loss of Power or Power control.	

62. If applicable, carry out a magnetometer interference test.
63. Carry out a magnetometer calibration and compass swing.
64. Carry out a flight check after satisfactory completion of ground testing. Check the alignment and brightness of the lights is satisfactory.

Part C – Weight and Balance

65. Determine the weight and location of the removed equipment and amend the aircraft weight and balance using the data from Table 8 or Table 9.

Table 8 : Installed Items (SI Units)

Component	Weight (kg)/Unit	Quantity	Arm (mm)	Moment
Landing and Taxi Light Assembly	0.96	2	1143	2,194.56
Position/Anti-Collision Lights (Wing)	0.24	2	1400	672.00
Position/Anti-Collision Lights (Vertical Fin)	0.15	1	7380	1,107.00
Fin Conduit	0.5	1	6500	3,250.00
Total	3.05		2368.38	7,223.56

Table 9 : Installed Items (Imperial Units)

Component	Weight (lb)/Unit	Quantity	Arm (in)	Moment
Landing and Taxi Light Assembly	2.11	2	45.00	190.31
Position/Anti-Collision Lights (Wing)	0.53	2	55.12	58.27
Position/Anti-Collision Lights (Vertical Fin)	0.33	1	290.55	96.00
Fin Conduit	1.10	1	255.91	281.83
Total	6.72		93.24	626.41

Documentation:

Update aircraft log book to reflect incorporation of this Service Bulletin.

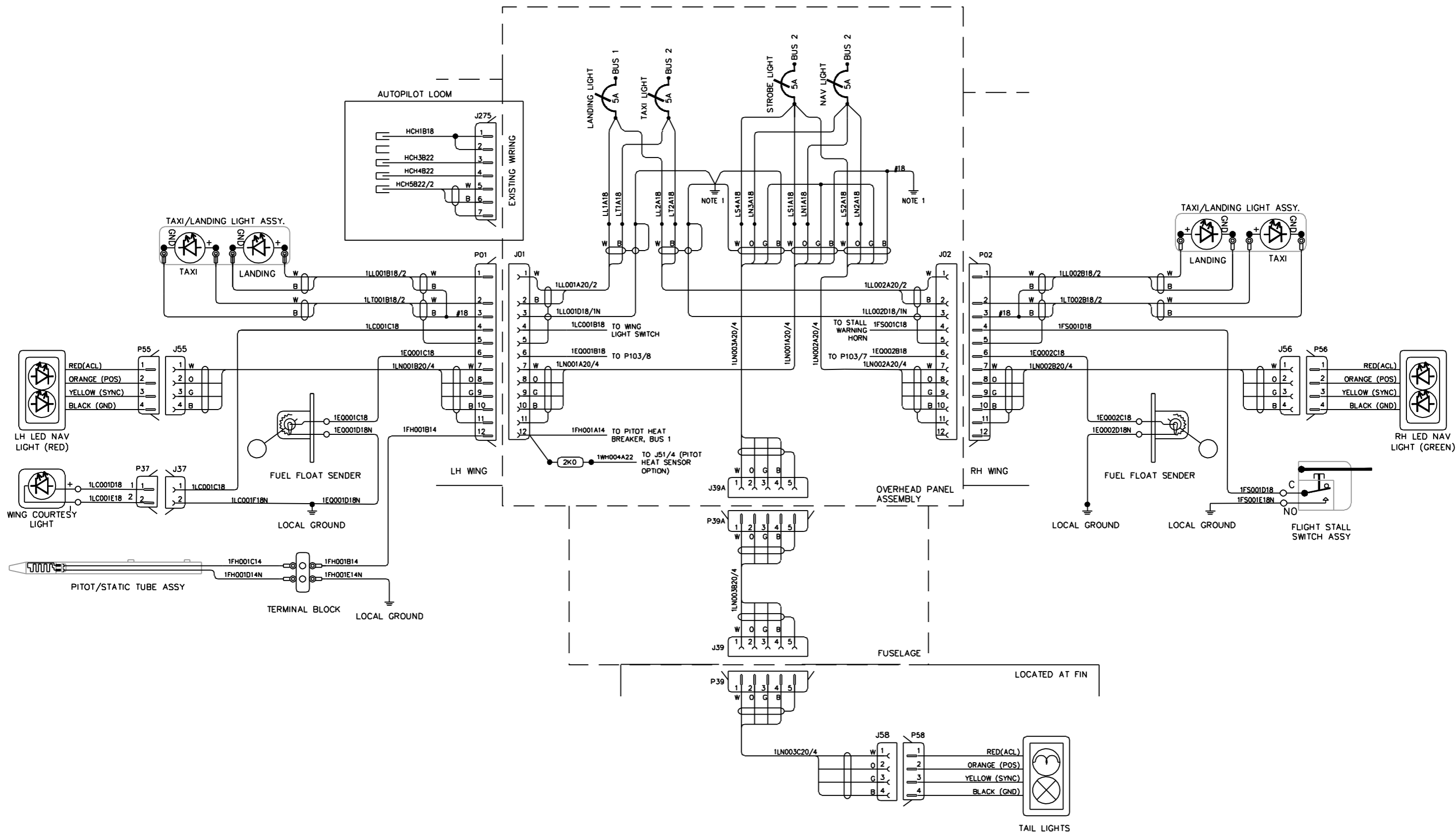
Insert this Service Bulletin and Service Manual Supplement, C05-96-36 Wing and Vertical Fin LED Lighting System ICA, Issue 17-Aug-2016 or later approved into the aircraft's Service Manual.

Continuing Airworthiness:

Instructions for Continued Airworthiness are contained in SMS C05-96-36

Compliance Notice:

Complete the Document Compliance Notice and return to GippsAero by mail, fax or email.



NOTE 1 : Refer Overhead Panel & Fuselage Wiring procedure for grounding point.

Figure 10 Wiring Schematic

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DOCUMENT COMPLIANCE NOTICE



A Mahindra Aerospace Company

Document:

SB-GA8-2021-207

Issue 1

Aircraft Serial Number: GA8-_____

Service Bulletin SB-GA8-2021-207, Issue 1 has been incorporated in the above aircraft.

Date of Incorporation: _____

Signed

Print Name: _____

If this Service Bulletin requires any inspections be carried out, describe the result of these inspections:

Please post, fax or email this compliance notice to:

GippsAero Technical Services
P.O. Box 881
Morwell Victoria 3840
Australia
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